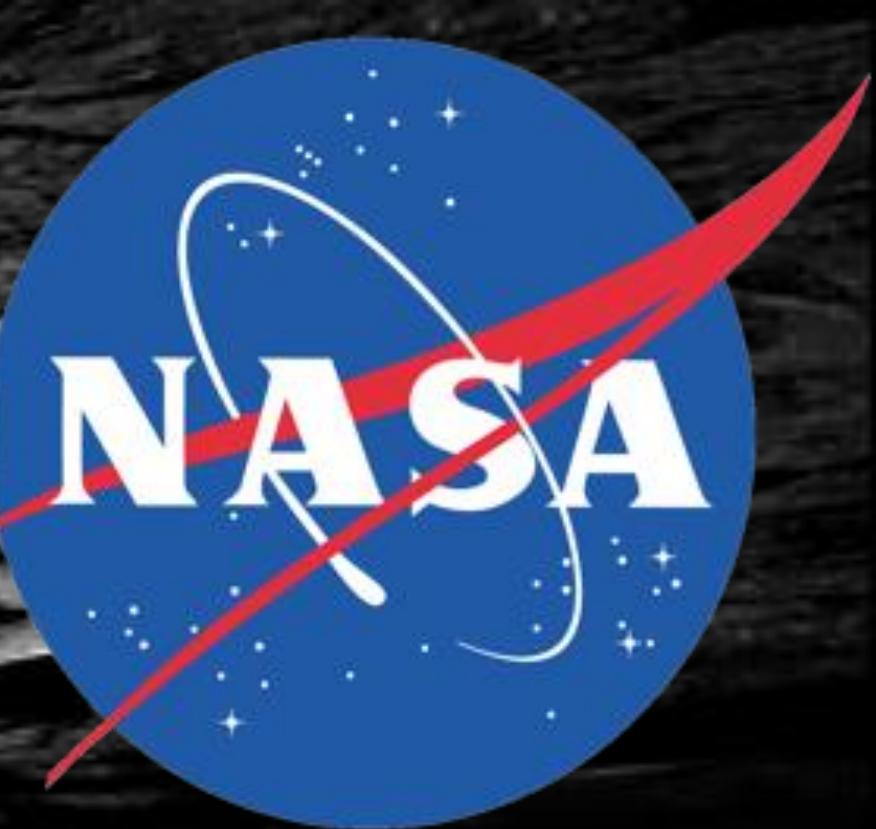


# System Performance Modeling of the Lunar Flashlight CubeSat Instrument



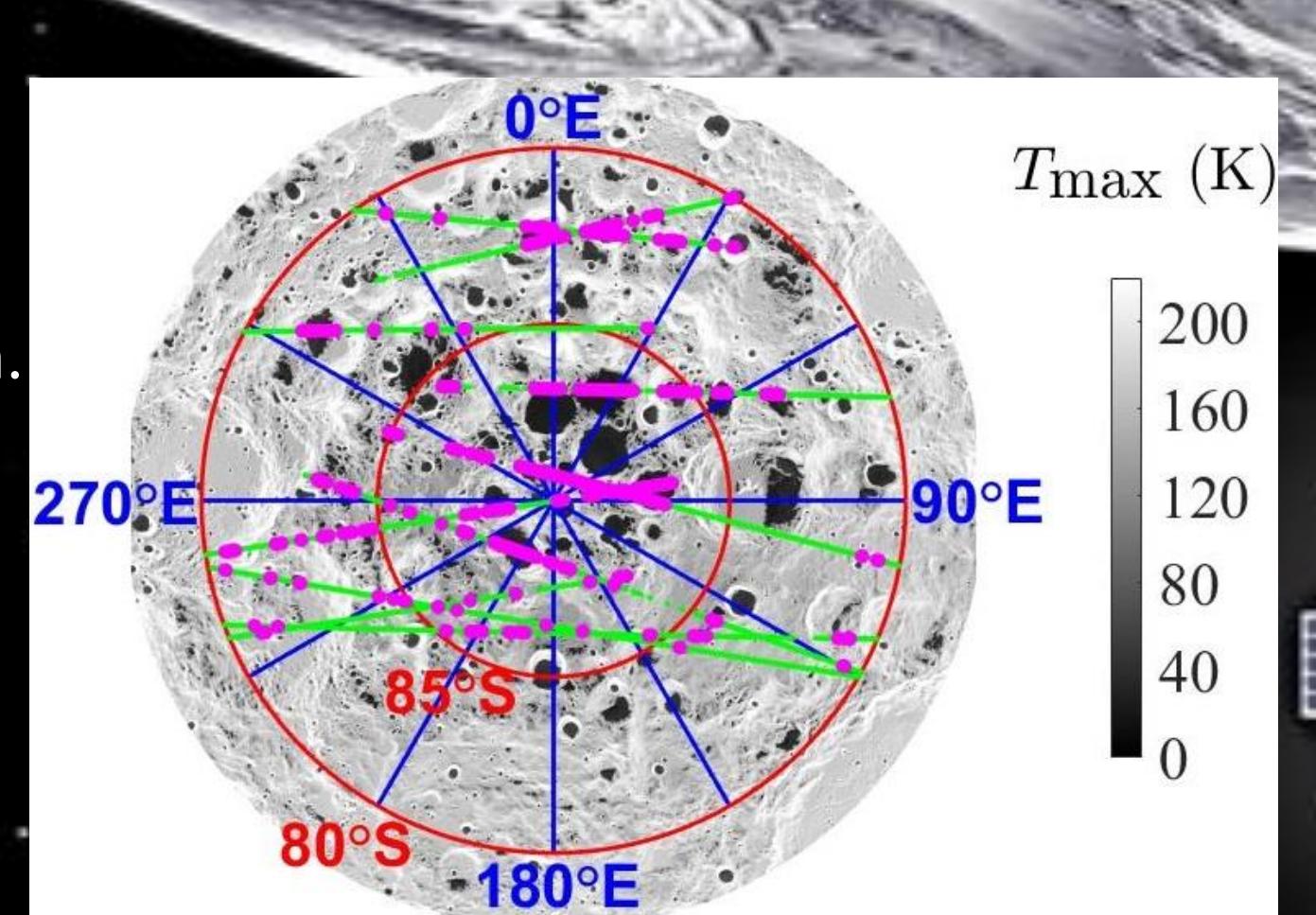
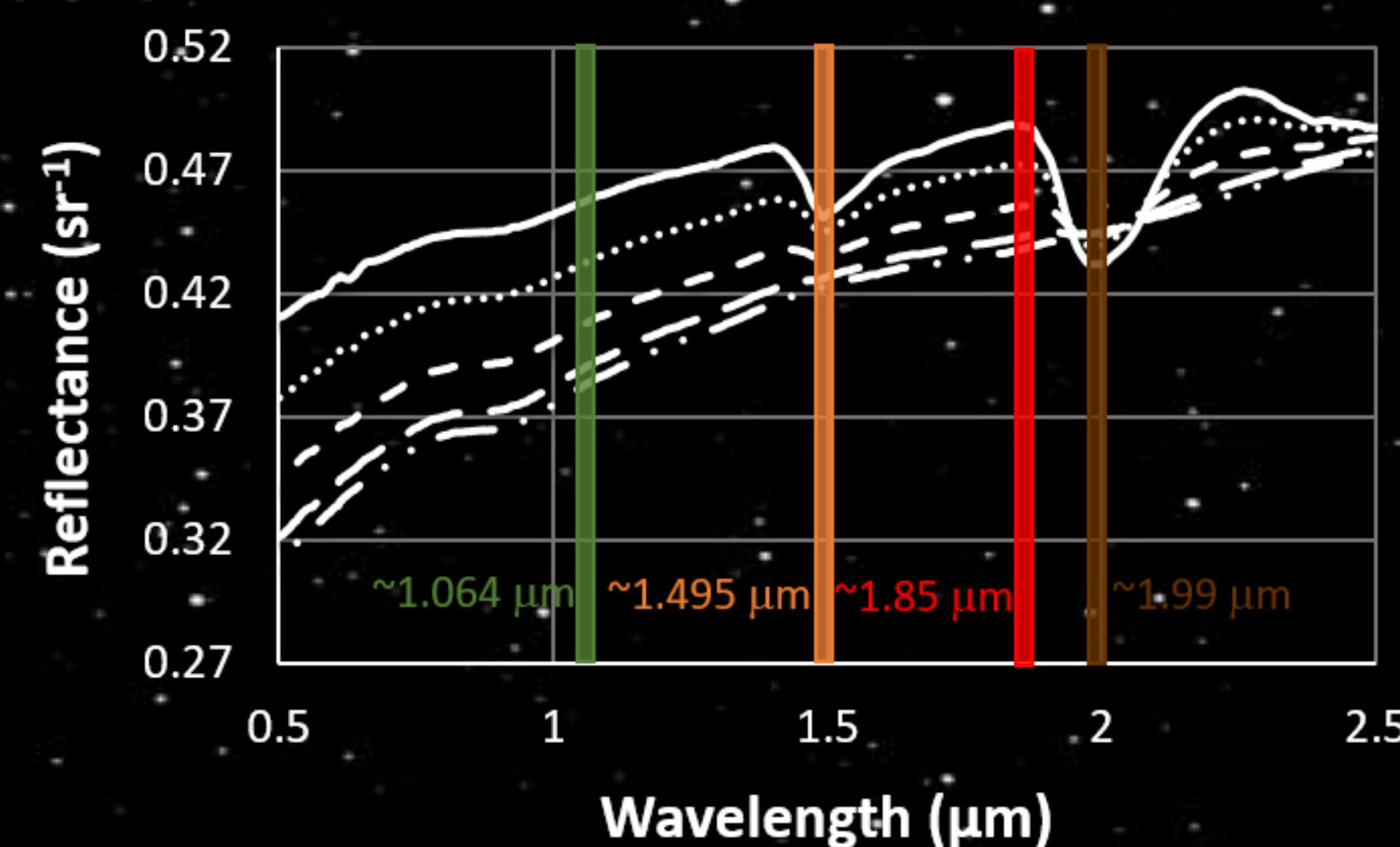
Quentin Vinckier (Jet Propulsion Laboratory, California Institute of Technology – vinckier@jpl.nasa.gov)

Paul O. Hayne (University of Colorado), Jose M. Martinez-Camacho, Christopher Paine, R. Glenn Sellar, Udo J. Wehmeier (Jet Propulsion Laboratory, California Institute of Technology), Barbara A. Cohen (NASA Goddard Space Flight Center)

## Lunar Flashlight (LF) mission

→ Goal ? Detect, quantify and map surface H<sub>2</sub>O ice in the PSRs (Permanently Shadowed Regions) – where previous missions have already revealed strong indications of H<sub>2</sub>O ice presence [1-6] – and occasionally sunlit regions of the lunar South Pole.

→ How ? By measuring 0° phase angle bidirectional surface reflectance band ratios using four different wavelengths in the SWIR (shortwave infrared) spectral region between 1 and 2 μm.



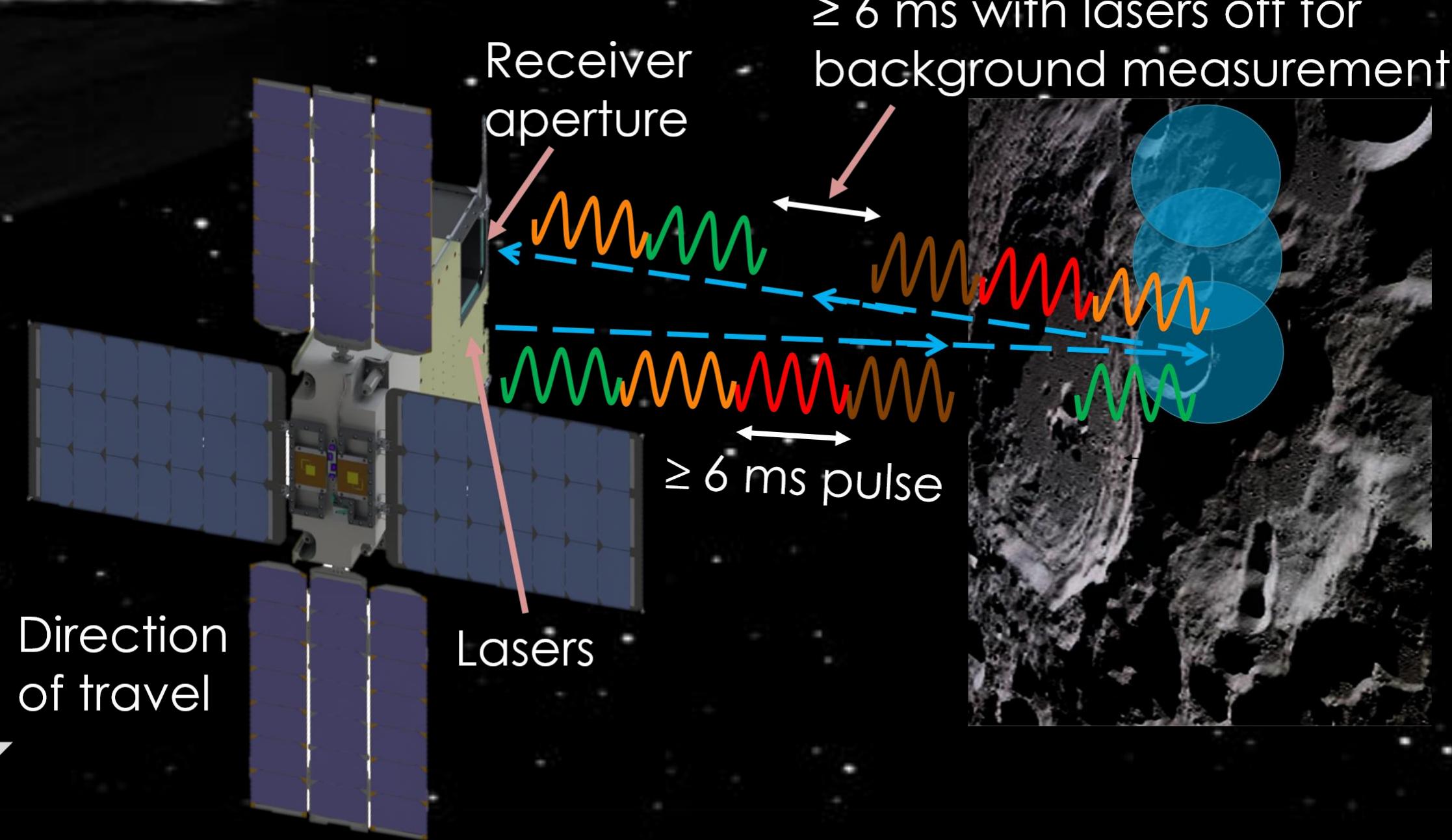
LF spacecraft Science data paths on the lunar South Pole. The gray scale shows the maximum temperature; the green and pink segments are the Science paths with the pink dots indicating the PSRs.

→ The lasers fire sequentially for  $\geq 6 \text{ ms}$ , followed by a pause of  $\geq 6 \text{ ms}$  with all lasers off.

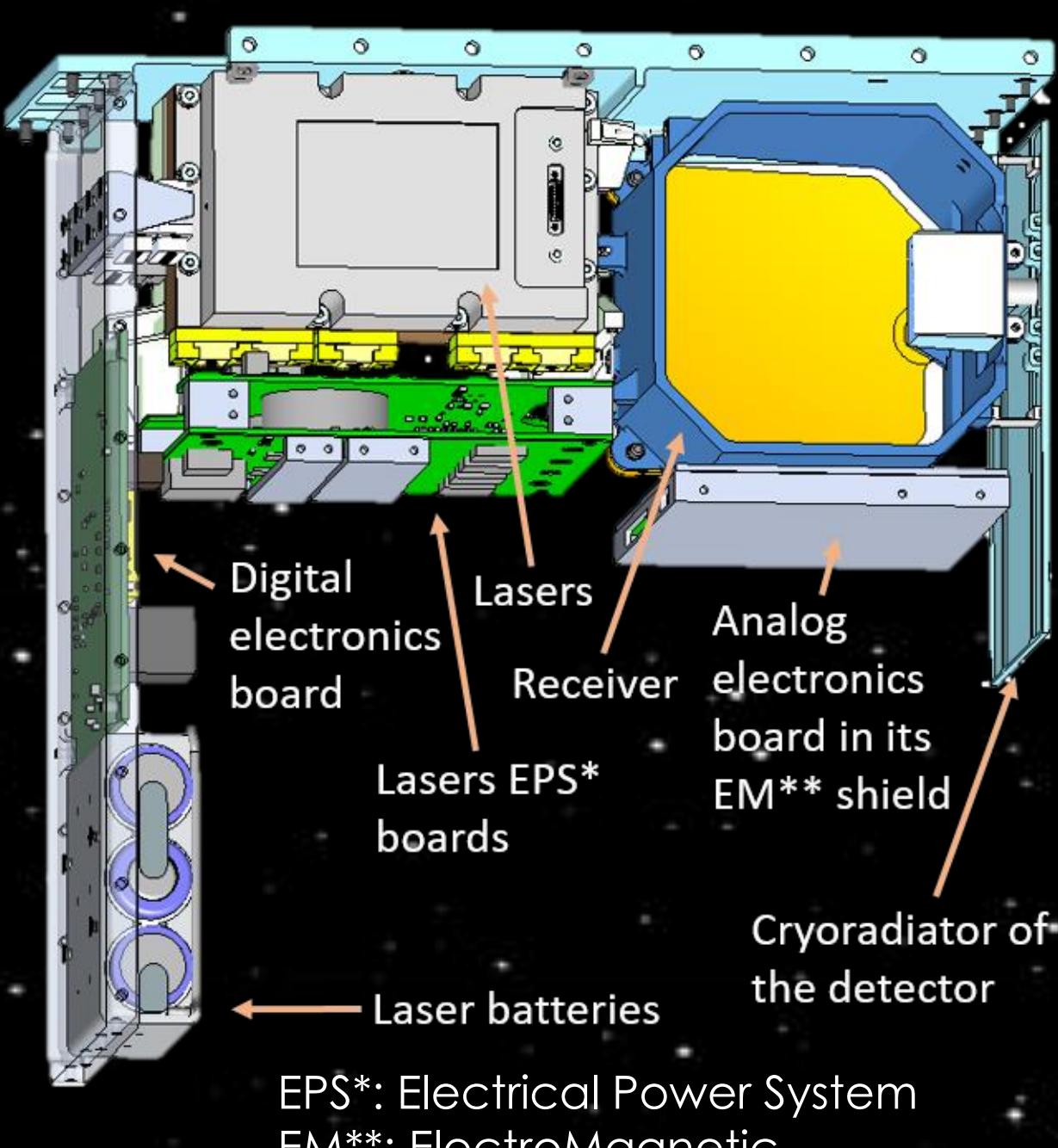
→ When all lasers are off, background is measured and subtracted from the measured signals.

→ In order to increase the SNR (Signal-to-Noise Ratio), measurements are averaged for each of the spectral band over the lunar ground-track corresponding to the desired mapping resolution.

→ Continuum/absorption reflectance band ratios are then analyzed to quantify the weight percent of water ice (wt%) in the illuminated FOV (Field Of View).



## LF multi-band SWIR laser reflectometer



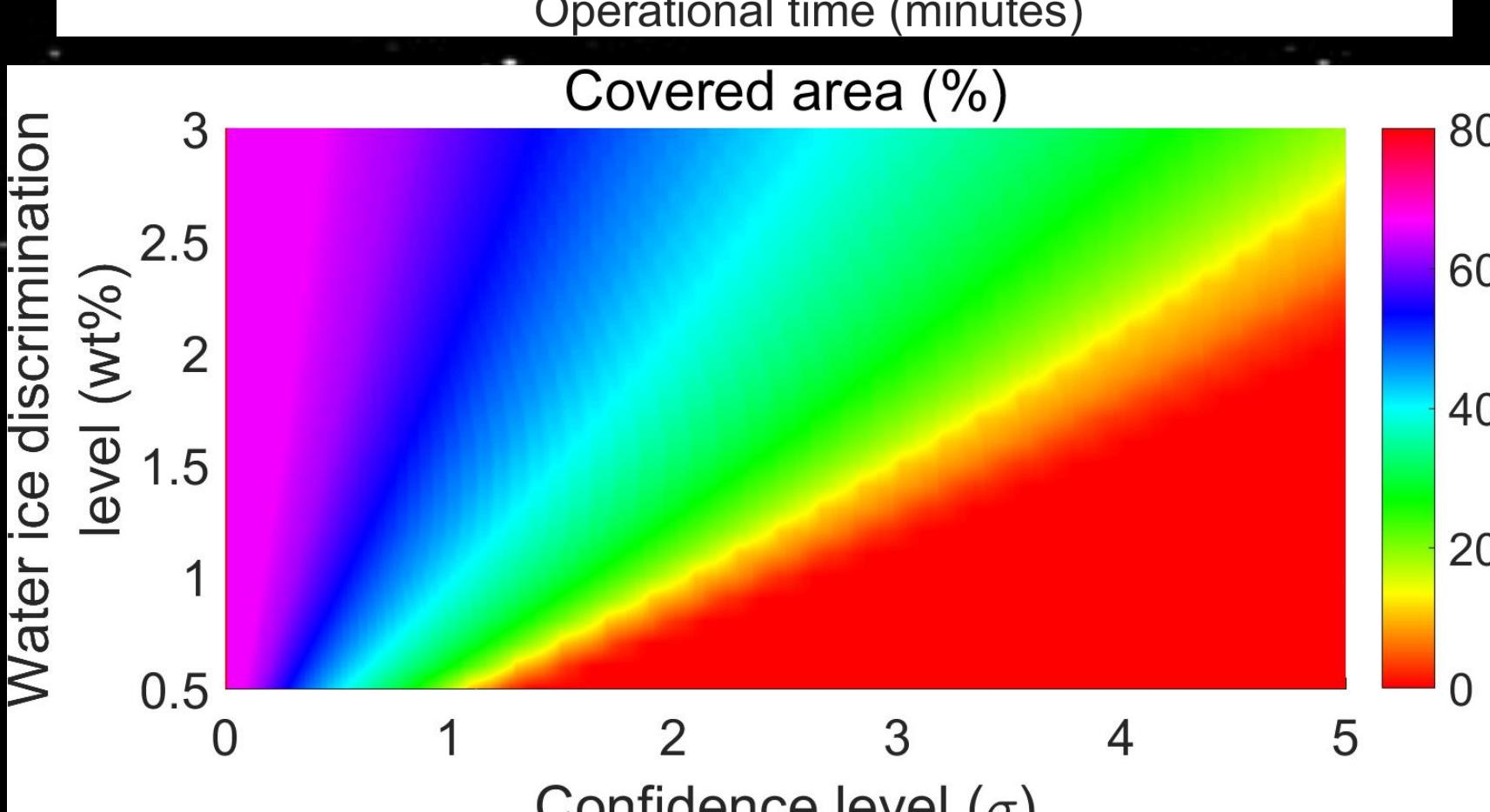
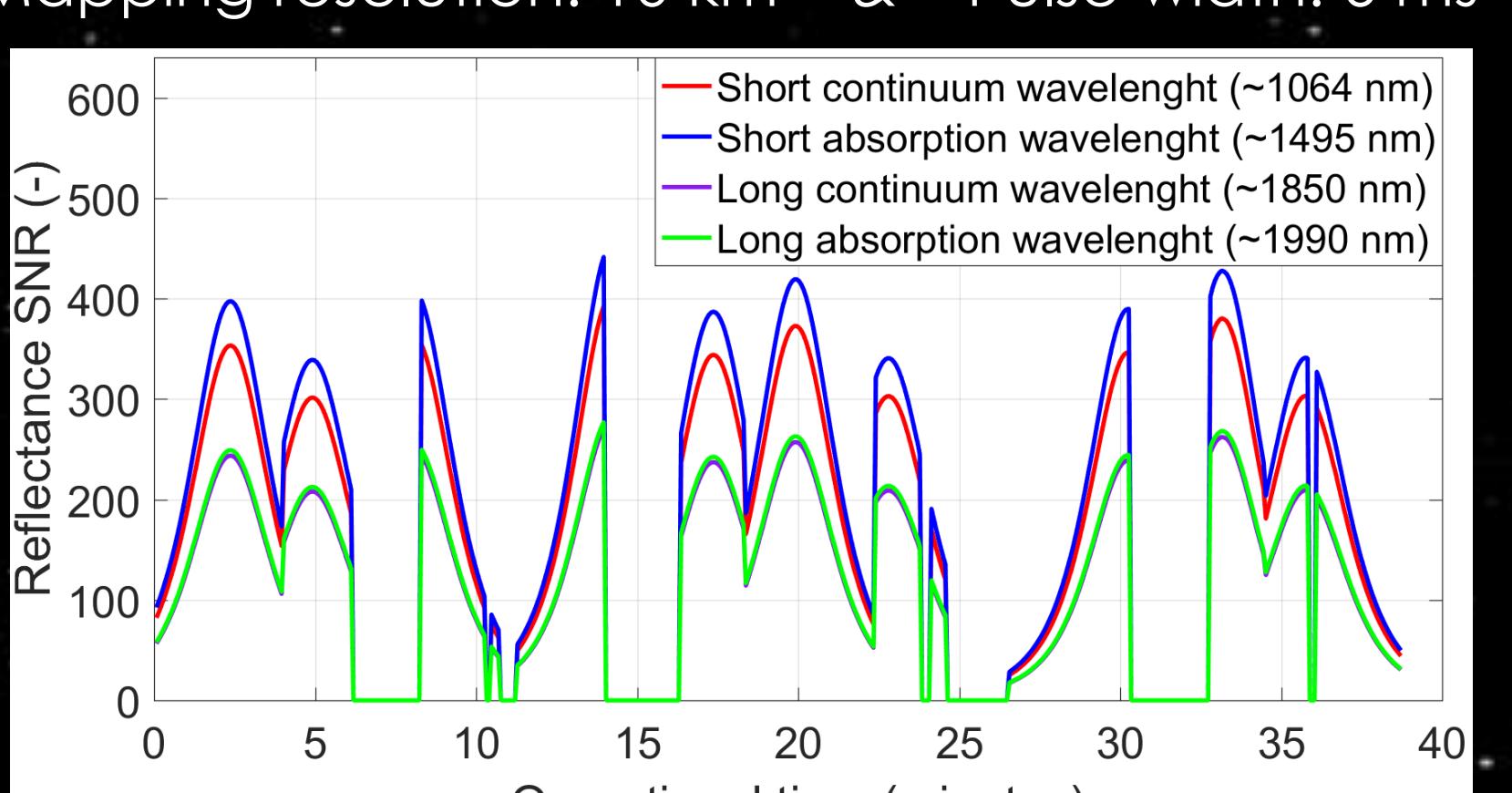
→ Optical receiver: 70×70-mm off-axis Al paraboloidal mirror; 70 mm focal length; single pixel 2 mm diameter InGaAs detector cooled at -65°C (1 nA dark current); 20.2 mrad uniform FOV; optimized for stray solar light rejection from outside the FOV.

→ Detector reading electronics (analog board): 0-10 nA current range; 100 kHz sampling rate; 0.5 pA/Hz<sup>1/2</sup> RMS (Root Mean Square) Nyquist noise spectral density; 236 Hz 3dB-bandwidth.

→ Diode lasers: 15-55 W optical power; wavelengths: ~1.495 & ~1.99 μm (H<sub>2</sub>O ice absorption peaks) and ~1.064 & ~1.850 μm (nearby continuum); > 99.6% emitted energy encircled within 20.2 mrad.

## Results and Discussion

Mapping resolution: 10 km & Pulse width: 6 ms



### Conclusion:

- The detector reading electronics (analog board) constitutes the major source of noise. Noise performance is mainly driven by the challenging mass and volume (2U) constraints of the instrument.
- Thermal constraints, driven by mass and volume constraints, limit the laser power available, and thus the instrument SNR.
- Trade off between mapping resolution and instrument SNR.

### Next steps:

- Finalize the system engineering phase to optimize the instrument SNR (lasers' power monitoring and pulse width VS background fluctuations & analog board bandwidth).
- Estimate TBD values in the error tree (achievable instrument calibration accuracy, background fluctuations impact, lasers-to-receiver pointing error) and finalize the calculations of the instrument performance.